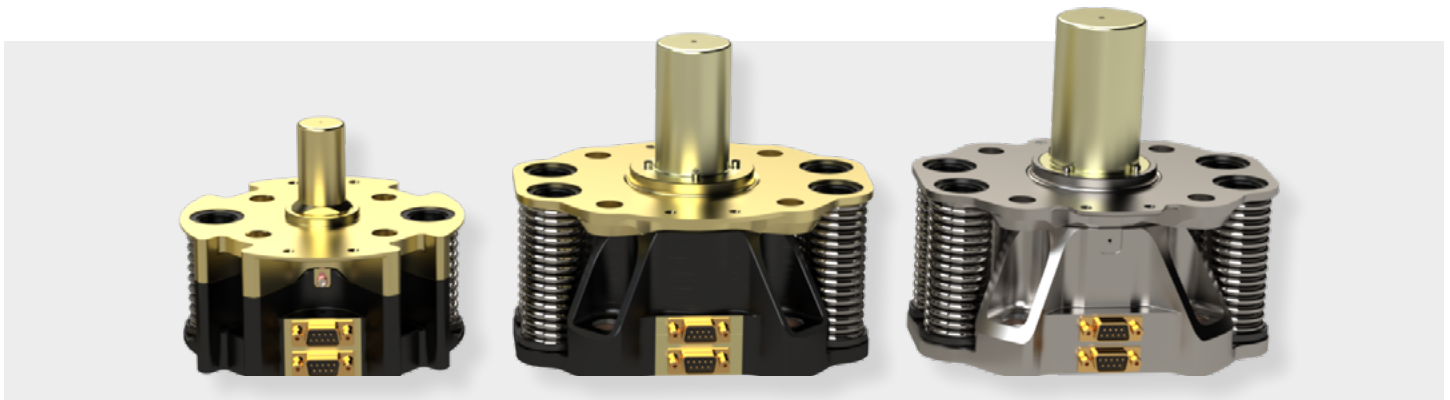


NEA® Payload Release Module

PRM™ 9103B, PRM™ 9104A, and PRM™ 9106A



The NEA® Payload Release Module (PRM™) is designed to mount, hold, and release a spacecraft from the launch vehicle upper stage, or from another carrying spacecraft (OTV for instance). The PRM™ is designed to be used in a multi-point release system for spacecrafts up to 2,000 kg in mass.

The PRM™ utilizes the flight proven NEA® 9103, 9104 or 9106 as the primary release mechanism, integrating a load carrying interface, separation springs, a bolt extractor, and optional separation switch or separation connector to offer a turnkey dispensing mechanism that is ideal for multi-point satellite release. The typical 4-point mount configuration of the PRM™ is designed to dispense payloads separating laterally or axially from a central dispensing structure with high simultaneity. The PRM™ is delivered fully preloaded, and no custom tools nor on-site training are required for assembly, making integration into space platforms simple and efficient. The PRM™ utilizes our low shock NEA® hold down and release mechanism (HDRM) with over 25 years of 100% flight success and over 15,000 actuations in space.

Principle of Operation

The PRM™ is an ultra-low shock, electrically initiated, single-shot, and factory refurbishable release mechanism that has the ability to carry a high tensile preload until commanded to release. The preload is applied through a release rod held in place by two separable spool halves which are in turn held together by tight winding of restraining wire. The restraint wire is held in place by redundant electrical fuse wires; actuation of either circuit allows release, assuring maximum reliability. When sufficient electrical current is applied, the restraint wire unwinds allowing the spool halves to separate releasing the release rod and the associated preload. The tunable separation springs provide the push off force to achieve the required separation velocity.

The PRM™ has a location to hold either a telemetry separation switch or 37 pin separation connector. The telemetry switch is commonly used to verify spacecraft separation from the launch vehicle, and the separation connector can be used for separation detection, ground power delivery or communications from the launch vehicle to the spacecraft.

Key Features

- Ultra-low release shock
- Can be operated with standard launch vehicle firing circuitry
- No debris generation
- High simultaneity of multiple hold-down points
- Customizable separation velocity
- Wide operating temperature range
- Factory refurbishment available
- NEA® HDRM with more than 25 years of flight heritage



Scan to view the
PRM™ 9103A animation.

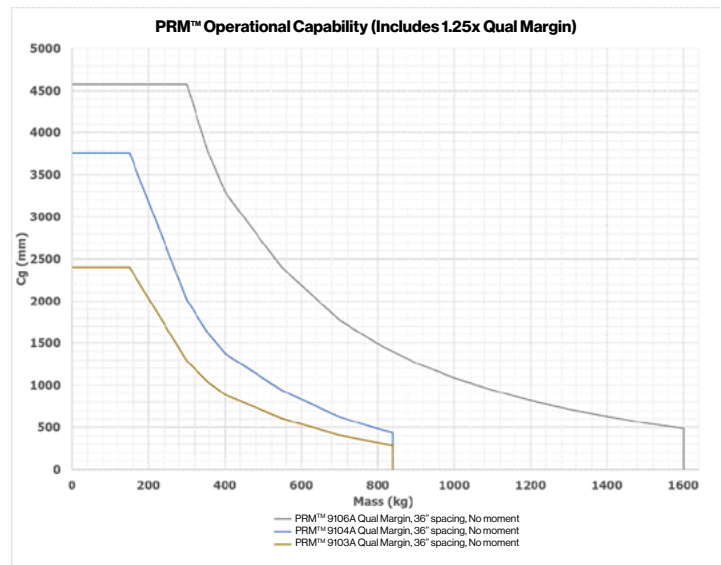
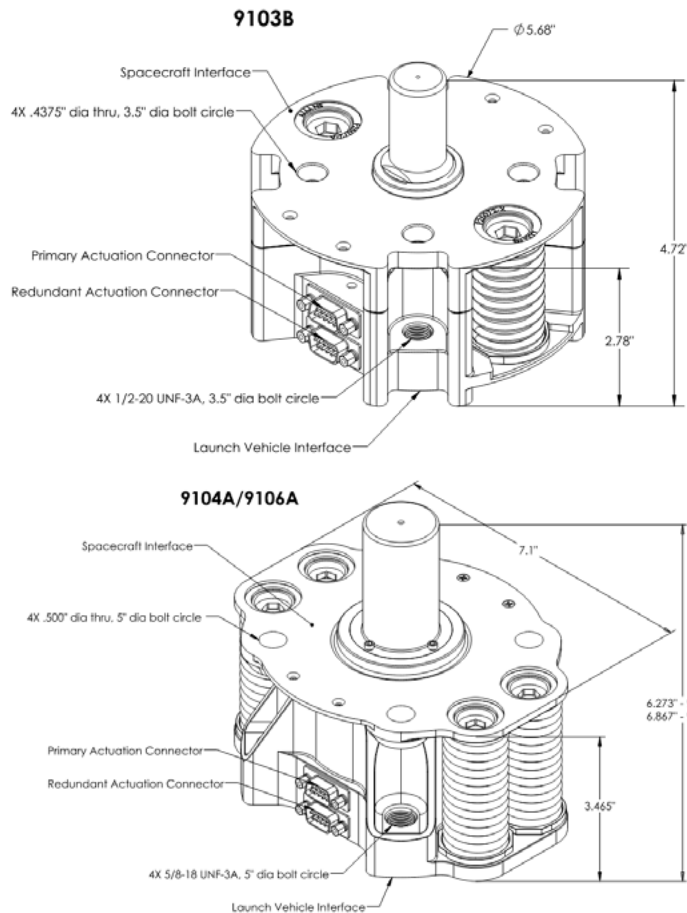


9103A Released

NEA® Payload Release Module – Technical Specifications

Parameter	PRM™ 9103B	PRM™ 9104A	PRM™ 9106A
Axial Preload	Up to 12,000 lbf	Up to 15,000 lbf	Up to 32,000 lbf
Axial Tension Stiffness	1.68E+6 lbf/in	8.8E+5 lbf/in	2.73E+6 lbf/in
Axial Compression Stiffness	4.76E+6	3.10E+6	8.5E+6
Shear Stiffness	1.64 E+6 lbf/in	1.34 E+6 lbf/in	2.54 E+6 lbf/in
Bending Stiffness	4.37 E+6 in-lbf/rad	1.61 E+6 in-lbf/rad	4.5 E+6 in-lbf/rad
Shock Output for Payload	< 350g's	< 500g's	< 750g's
Firing Current	≥ 6 or ≥ 3 amps	≥ 6 or ≥ 3 amps	≥ 6 or ≥ 3 amps
Firing Duration	35 or 50 ms	35 or 50 ms	35 or 50 ms
Release Time	< 60 ms	< 80 ms	< 80 ms
Simultaneity	< 6 ms	< 10 ms	< 15 ms
Total Mass	< 3.3 lbm (w/ 2 springs)	< 6.5 lbm (w/ 2 springs)	< 11.8 lbm (w/ 4 springs)
Flyaway Mass	< 1.17 lbm	< 1.8 lbm	< 5.2 lbm
Temperature Range	-40°C to +101°C	-40°C to +101°C	-40°C to +101°C
Nominal Spring Energy(J) (Configurable)	2.8 to 16.2 Joules	3.3 to 40.0 Joules	3.3 to 40.0 Joules

Mechanical Interface Drawings



*Includes 1.25x qual margin to F9 RPUG environments margin to SpaceX Falcon 9 Rideshare Payload User Guide Environments. Contact EBAD for cabalilities on other launch vehicle provider environments.



Attention: The information and recommendations described in this brochure cannot possibly cover every application of the products or variation of conditions under which the products are used. The recommendations here in are based on the manufacturer's experience, research and testing. They are believed to be accurate, but no warranties are made, express or implied. In addition, the specifications contained herein are all nominal which represent our current production. The products described may be subject to change. Please feel free to contact Ensign-Bickford Aerospace & Defense Company for verification. No Warranties or Liabilities: The products described herein are sold "AS IS" and without any warranty or guaranty, express, or implied, arising by law or otherwise including without limitation any warranty of merchantability or fitness for a particular purpose. Buyer and user agree further to release and discharge seller from any and all liabilities whatsoever arising out of the purchase or use of any product described herein whether or not such liability is occasioned by seller's negligence or based upon strict products liability or upon principles of indemnity or contribution. Content©2023 Ensign-Bickford Aerospace & Defense Company, Simsbury, CT 06070, U.S.A.